

Computational studies of pressure distribution around a bending 5:1 rectangular cylinder

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SUMMARY:

This paper describes the development and testing of a numerical model, based around the Large Eddy Simulation model, of a flexible 5:1 rectangular cylinder. We report on the vortex-induced vibration (VIV) of the bending cylinder, which is an analogue for a flexible bridge deck. Compared with a 3D heaving sectional model, distinct differences in surface pressure distributions and Strouhal number in the span-wise direction were found. In addition, certain span-wise flow features were thought to trigger the lock-in when the flexible model underwent the bending VIV response.

Keywords: Bridge aeroelasticity, 5:1 ratio, Rectangular cylinder, Pressure distribution, CFD, Fluid-structure interaction.

1. INTRODUCTION

The rectangular cylinder has been considered as a simplified geometry of many structures in the built environment including the main section of a suspension bridge. It differs from the circular cylinder in that the rectangular cylinder is characterised by the permanent separation at the leading edges causing two unstable shear layers along surfaces of the body. The aspect ratio (i.e. the ratio between the width and the depth of the cross section) and the Reynolds numbers have been found to control their properties together with the vortex structure in the wake region (Okajima, 1982; Ozono et al., 1992; Schewe, 2013). More importantly, the interaction between the leading-edge shear layer and the von Karman vortex shed from the trailing edge governs the dynamic responses (Kotmasu and Kobayashi, 1980; Matsomoto et al., 2008).

In this paper, the 5:1 rectangular cylinder is selected as a case study since it is representative of a generic bridge deck geometry. Wind tunnel tests performed by Stokes and Welsh (1986) and Mills et al. (2003) indicated this section is classified into the category of intermittent reattachment flow where the aforementioned shear layers form two separation bubbles reattaching at points very close to the trailing edge. The geometry of these bubbles was found to be affected by the Reynolds number and especially by the turbulence in the flow (Kiya and Sasaki, 1982). Matsumoto et al. (2003) pointed out the turbulence-induced effects on the pressure distribution around a 5:1 rectangular cylinder, including a decrease in the span-wise pressure correlation as the turbulence level increases. Later, Haan and Kareem (2007) suggested this alteration of the surface pressure is responsible for the damping effects of turbulence on flutter which was previously observed by Scanlan (1997). In addition, Wu and Kareem (2012) showed that increased turbulence produces a stabilising effect on the VIV of a 5:1 rectangular cylinder; however, a comprehensive explanation of the mechanism is still to be found.

The aerodynamics of the flow around and the aeroelasticity of the rectangular cylinder have also been investigated by the use of numerical methods. Due to the complexity of the problem and the limitation of computational power, the simulation was initially restricted to model the flow field around 2D static cylinder only with the use of the two-equation Unsteady Reynolds-Averaged Navier-Stokes (URANS) models. Later, 3D simulations using the LES model have become more available focusing on uncovering the characteristics of the separation bubbles, the effect of the afterbody length on the separation and reattachment of the flow and the coherence structure of the surface pressure around a static rectangular cylinder (Bruno et al., 2010). The LES approach was also coupled with a structural solver to model the fluid-structure interaction (FSI) of a 3D elastically supported rectangular cylinder at a low Reynolds number flow (Sun et al., 2008). The work highlighted the suitability of the LES model to capture the inherent unsteadiness in FSI problems and to maintain the turbulence structure in the wake region in contrast to over-dissipation of the URANS turbulence models. Daniels et al. (2004) also applied this method to predict the effect of the free-stream turbulence on the aerodynamic behavior of a static and elastically mounted 4:1 rectangular cylinder. The use of the LES approach is still limited due to its computationally expensive near-wall treatment. In addition, Bruno et al., (2011) showed results of 3D LES simulations in the application of the bridge aeroelasticity were very susceptible to the level of discretisation in the span-wise direction.

In this project, utilising the High Performance Computer at the University of Nottingham, the aforementioned coupling approach between the LES model and the FSI solver is applied to model the flow field and simulate the aeroelasticity of a static and elastically mounted 3D sectional rectangular cylinder of the 5:1 aspect ratio in the smooth flow. In addition, a 3D flexible model is built and a mode shape is integrated in the FSI solver to simulate its responses as a half of the bridge deck. The surface pressure distribution and the VIV responses of the flexible model in the bending mode will be compared against those of the sectional model in the heaving mode to uncover the important flow features in the span-wise direction.

2. APPLICATION

The computational studies were calculated using the open-source CFD software OpenFOAM v2.2.2. The unsteady flow around the rectangular section was modelled using a LES model. The Smagorinsky model was applied to model sub-grid scale stresses. The computational domain and the key boundary conditions are summarised in Figure 1a. A zero gradient condition for velocity and constant value of zero pressure were imposed on the outlet. As for the inlet, a non-zero x-component wind speed and a zero gradient condition for pressure were specified simulate the smooth flow. The `movingWallVelocity` was applied on the surface of the model to accurately model a zero normal-to-wall velocity component particularly in the dynamic simulation. The computational domain was hybrid in the x - z plane (Figure 2a) and structured in the y direction. An inflation layer, which was a six-cell-thick structure grid, was imposed around the model. The discretisation next to the model was $\Delta x/B = 2\Delta z/B = 4 \times 10^{-3}$ (Figure 2b). Different values of the span-wise discretisation, Δy , span-wise length, L , and boundary conditions of the y patches between static and dynamic simulations were used and summarised in Table 1. The 3D sectional model was used to perform static and heaving simulations while bending simulations were carried out on the 3D flexible model where the deformation of the cylinder was modelled.

2.1. Static Simulation

The OpenFOAM solver `pimpleFoam` was used to simulation the flow around the 3D static sectional model at 3 different wind speeds, 1, 2 and 4 ms^{-1} . At each wind speed, the surface pressure data was extracted at selected points to evaluate the pressure correlation and pressure distribution.

2.2. Dynamic Simulation

As for the 3D sectional model, the coupling scheme procedure between the OpenFOAM solver `pimpleDyMFoam` and the mass-damper-spring equation was utilised. The mode shape equation is set as $\Phi_1(y) = 1$, i.e. the 3D section model behaved like a rigid body having a mass per unit length of $\bar{m} = 4.7 \text{ kg/m}$. The Scruton number was $S_{cr} = 9.64$. The model was restrained to respond in the heaving mode only with the natural frequency of $f_{n,h} = 1.2 \text{ Hz}$. The wind speed was increased from 0.1 ms^{-1} to 2.5 ms^{-1} in increments of 0.1 ms^{-1} during the lock-in.

For the 3D flexible model, the mode shape $\Phi_1(y) = \Phi_0 \sin[\pi/(2L_o) y]$ with the scaling factor $\Phi_0 = 0.363$ was used so that the model was treated as a cantilever which could exhibit half of the first bending mode (Figure 3). The selection of this scaling factor allowed a unit generalised mass. All of the structural parameters were similar to the 3D sectional model. It is noted the inclusion of the L_1 and $L_{abutment}$ in Figure 3 was to reduce the effect of the `symmetryPlane` boundary condition on the flow field around the half-span L_o section.

As for the dynamic simulation, a dynamic mesh algorithm was prescribed and implemented to accommodate the deformation of the model. As shown in Figure 1b, the computational domain is divided into 9 different blocks. Blocks 1 and 3 are rigid where cells are fixed relatively to the model. The other blocks are grouped into the buffer zone where cells are deformed and displaced to facilitate the displacement of the model. In addition, the conventional serial staggered algorithm is applied to model the coupling between the fluid, structure and dynamic mesh.

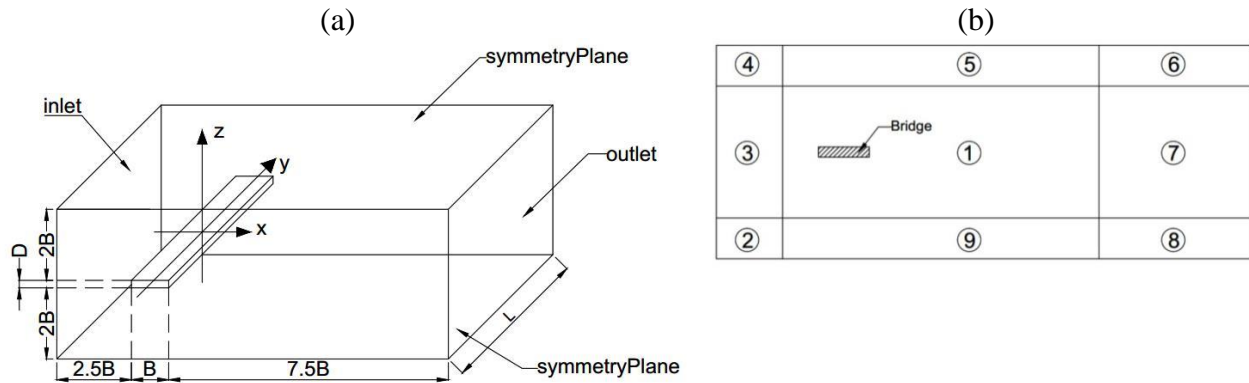


Figure 1. Computational domain and boundary conditions (a) and illustration of 9 different blocks in the computational domain (b).

Table 1. Span-wise discretisation, number of cells and boundary condition of y patches in three simulations.

Simulations	L/B	$\Delta y/B$	Number of layers	Number of cells	Boundary condition of y patches
3D static simulations	3	0.02	150	10.5 million	cyclic
3D heaving simulations	3	0.1	30	2.1 million	cyclic
3D bending simulations	7	0.1	70	4.9 million	symmetryPlane

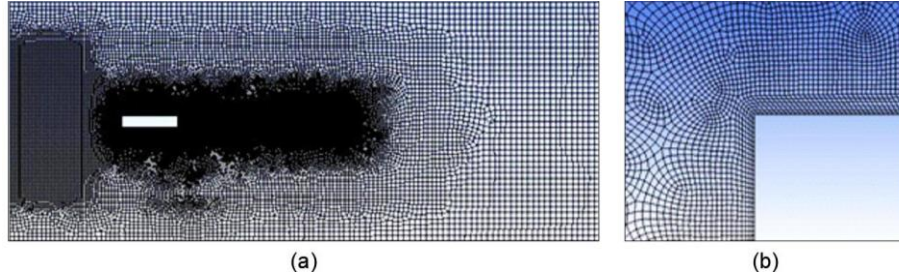


Figure 2. Computational grid in the x - z plane for the entire domain (a) and around the leading edge (b).

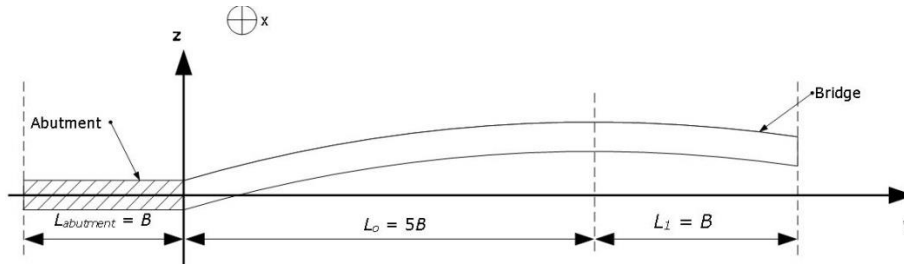


Figure 3. Schematic diagram of the simulated flexible model. $L_o = 5B$ is the half of the span.

3. RESULTS AND DISCUSSION

3.1. Static Simulation

Table 1 summaries force coefficients and Strouhal number at three different Reynolds numbers in a comparison against the on-going wind tunnel study (Nguyen et al., 2016) and another wind tunnel study (Schewe, 2013). The width, B , was used as the characteristic length scale for all coefficients. Except for the values of C_L , there was a good agreement between the current computational study and other wind tunnel tests. This study predicted a negative value of C_L , which was also reported in other LES simulations (Bruno et al., 2010). One possible reason for this discrepancy was the slight asymmetry between the top and bottom halves of the unstructured grid which may lead to the flow being resolved differently on either side of the model.

The flow field around the static model was investigated by calculating the pressure correlation and coherence in the span-wise direction around the leading edge ($x/B = 0.18$) and the trailing edge ($x/B = 0.82$) at the Reynolds number of 6700. As shown in Figure 4, the pressure correlation around the leading edge was found to be higher than that around the trailing edge. Selected wind tunnel study also showed similar behaviour despite the difference in the Reynolds number and the effect of the `cyclic` boundary condition. This behaviour is thought to be due to the presence of the well-defined separation bubble along the span-wise length compared to the intermittent flow feature at the reattachment point close to the trailing edge. This result could also be inferred from the coherence structure of the surface pressure (Figure 5). In addition, these graphs also indicate the dominant fluctuation of the surface pressure inside the separation bubble and at the reattachment point occurred at the vortex shedding frequency given by the Strouhal number. Therefore, as for the 5:1 rectangular cylinder, the vortex shedding and the flow field around the section was governed by the synchronisation of the shear layer created from the flow separation at the leading edge and the vortex shed from the trailing edge into the wake region.

Table 2. Force coefficients and Strouhal number obtained from computational and wind tunnel studies.

	Re	St	C_D	C_L	C'_L
Current CFD studies	6700	0.608	1.205	-0.056	0.081
	13000	0.600	1.030	-0.059	0.075
	27000	0.609	1.030	-0.063	0.059
WT studies - Schewe (2013)	6000 – 40000	0.555	1.210	~0	~-0.08
WT studies - Nguyen et al. (2016)	50000	0.640	1.191	0.034	-

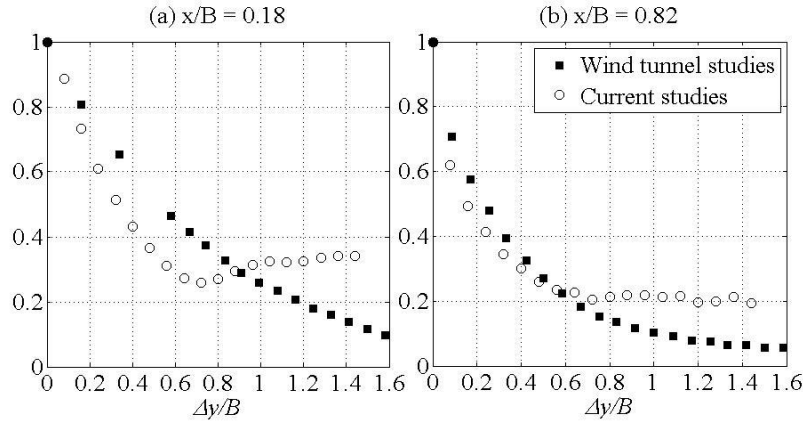


Figure 4. Pressure correlation in the span-wise direction. Wind tunnel results from Ricciardelli and Marra (2008) at the Reynolds number of $Re = 63600$.

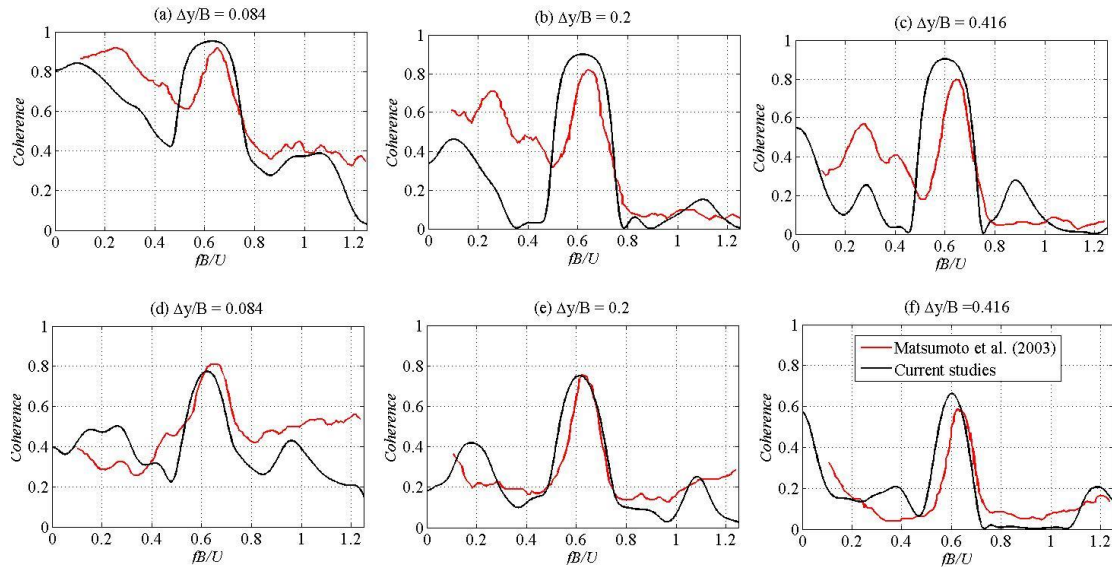


Figure 5. Span-wise coherence of pressure. The top row is close to the leading edge while the other is close to the trailing edge.

3.2. Heaving Simulation

The behaviour of the sectional model restrained to heaving only in the smooth flow is illustrated in Figure 6. Two VIV responses were observed at reduced wind speeds of $U_r = 0.83$ and 1.67 (Figure 6a). They were the two lock-in regions indicated by an increase in the amplitude of the response and the fact that the vortex shedding frequency was locked into the heaving natural frequency (Figure 6b). They are the two harmonics of the heaving VIV response. In addition,

Figure 6c showed the variation of the phase of the lift force against the displacement with the wind speed. As the amplitude of the response increased, the in-phase component of the lift force became less and when the cylinder moved out of the lock-in region, the lift force suddenly became out-of phase. This transition indicated there was a dramatic change in the flow structure around the cylinder which was responsible for the lock-out.

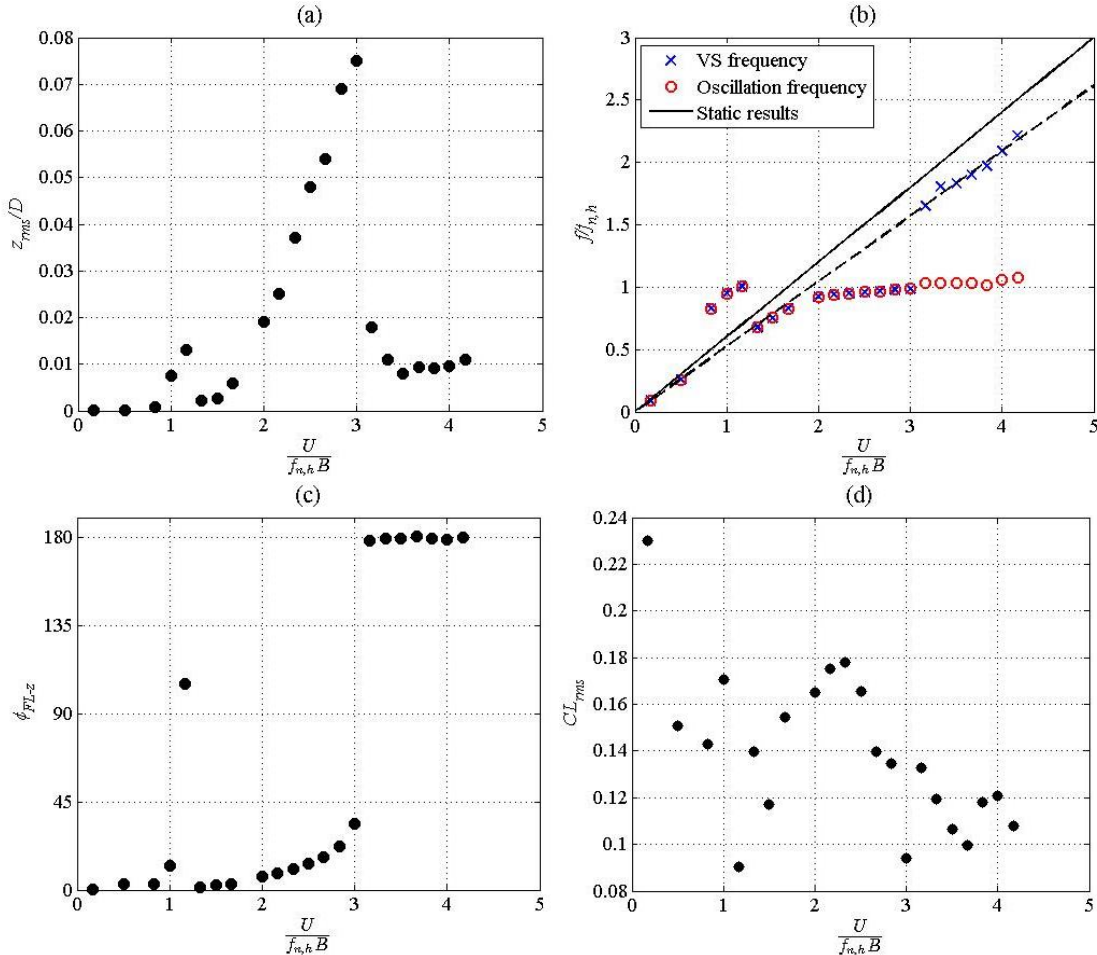


Figure 6. Heaving VIV response of the sectional model in the smooth flow.

An investigation of the span-wise correlation of the surface pressure revealed a significant variation in the flow field around the cylinder as it underwent the lock-in. At $U_r = 1.67$ which was the onset of the lock-in, the amplitude of the response was small and the flow field around the cylinder shared some similar features as the static cylinder as shown in Section 3.1. When the amplitude of the response increased ($U_r = 2.00$ to 2.67) and reached the peak ($U_r = 3$), a slight decrease in the correlation level around the leading edge was observed while around the trailing edge, the flow field was better correlated. As the lock-out occurred, the correlation level around the trailing edge suddenly dropped. Similar results were also found in our on-going wind tunnel studies. It indicated that, particularly for this 5:1 rectangular cylinder, the motion-induced leading edge vortex acted as a triggering mechanism for the VIV response while a strongly correlated flow feature occurred around the trailing edge, which made the amplitude rise. It could be the impingement of the motion-induced leading edge vortex on the surface of the cylinder resulted from the interaction of the flow field and the motion of the cylinder.

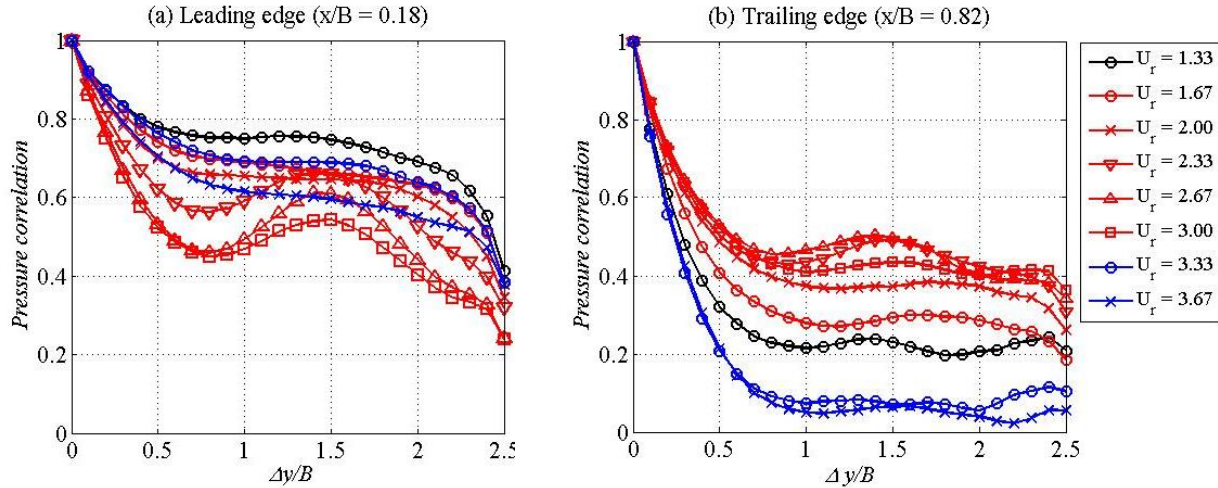


Figure 7. Variation of the span-wise pressure correlation around the leading and trailing edges as the cylinder experienced the lock-in.

3.3. Bending Simulation

As can be seen in Figure 8, the VIV response at the mid-span ($y/B = 5$) of the flexible model shared some common characteristics as the sectional model regarding the onset reduced wind speeds of two VIV lock-in regions, the lock-in of the vortex-shedding frequency and the reduced wind speed where the maximum response occurred.

At the reduced wind speed of $U_r = 3$ which was at the peak of the larger VIV response, the pressure distribution measured at different span-wise positions is shown in Figure 9; s is measured from the stagnation point on the front face. In contrast to the heaving model where flow features on the side surfaces were uniform in the span-wise direction, the geometry of the separation bubble varied significantly from the fixed end ($y/B = 0$) to the end having the maximum displacement ($y/B = 5$). Also the frequency of the lift force shifted from the value defined by the Strouhal number (Peak A in Figure 10a and b) to the natural frequency (Peak B in Figure 10b and c).

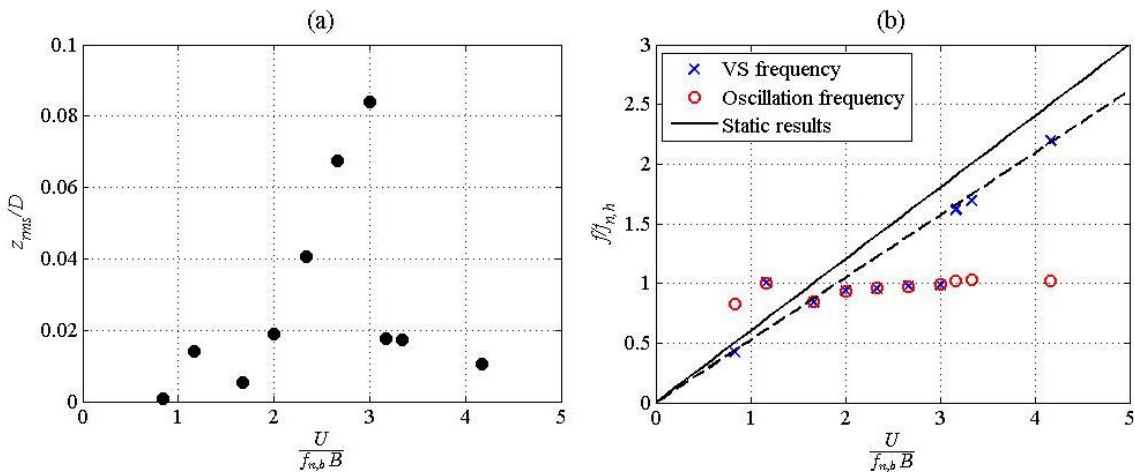


Figure 8. VIV response of the flexible model restrained to the bending mode only in the smooth flow; the displacement in (a) was measured at the mid-span ($y/B = 5$).

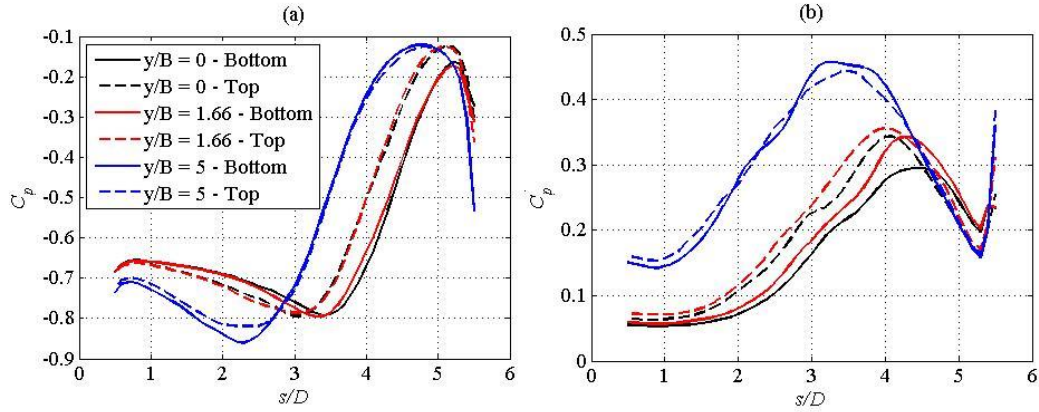


Figure 9. Pressure distribution on the top and bottom surfaces at three different span-wise positions.

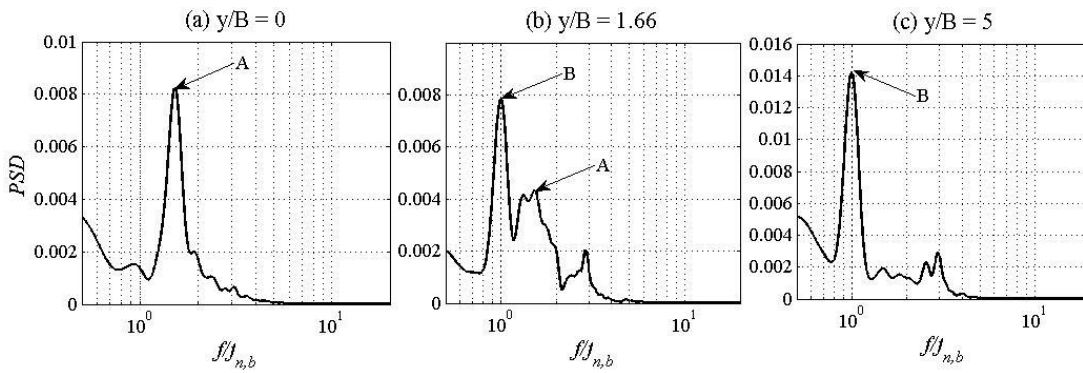


Figure 10. Spectra of the lift forces measured at different span-wise positions.

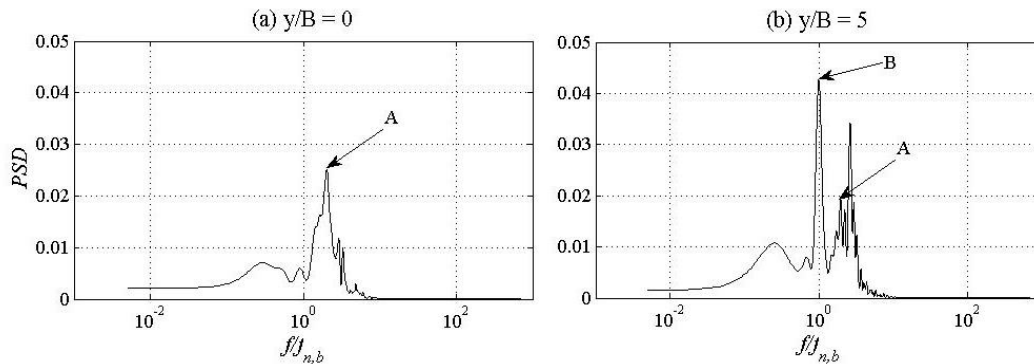


Figure 11. Spectra of the z -component velocity measured at a distance B behind the model and $D/2$ from the top surface.

The velocity measured behind the model also showed a variation in flow features in the span-wise direction (Figure 11). At the fixed end, the dominant flow feature was the von Karman vortex shed from body at the frequency defined by the Strouhal number (Peak A in Figure 11a). Towards the end having the maximum displacement, the motion-induced vortex became dominant (Peak B in Figure 11b); the flow feature found at the static end was still present but their relative strength varied along the span-wise length.

Figure 12 and Figure 13 show the variation of the phase angle of the lift force against the displacement and the value of C_L' in the span-wise direction as the flexible model experienced

the VIV lock-in. Around the moving end ($y/B = 5$), the general behaviour was very similar to what observed on the sectional model, which was a sudden change in the phase angle at $U_r = 3.33$ indicating the lock-out and the fluctuating lift force was maximum at $U_r = 2.33$. However, in the span-wise direction, some distinct flow feature was observed. At the onset of the lock-in which was around $U_r = 2$, regarding the phase relationship, the lift force was behind the displacement at the moving end ($y/B = 5$). Also, the span-wise profile of C_L' at this reduced wind speed showed less difference for the portion of $y/B > 1$. When the amplitude increased, this difference was magnified; more fluctuation in the lift force was observed around the moving end. The phase profile showed the lift force was in-front of the displacement and the in-phase component of the lift force reached the peak at $U_r = 2.33$ and gradually decreased. Eventually, a sudden change in the gradient of the span-wise profile of the phase lag indicated the lock-out.

These results showed that the on-set of the lock-in potentially did not relate to the aerodynamics of the flow field around the moving end where the maximum displacement is expected. Instead, some flow feature occurred around the portion between $y/B = 1$ and 3 could be the triggering mechanism for the lock-in. As the amplitude of the response increased, the flow feature was either diffused by the pressure gradient occurring in the span-wise direction or shifted towards the moving end and accounted to building up the amplitude of the VIV response.

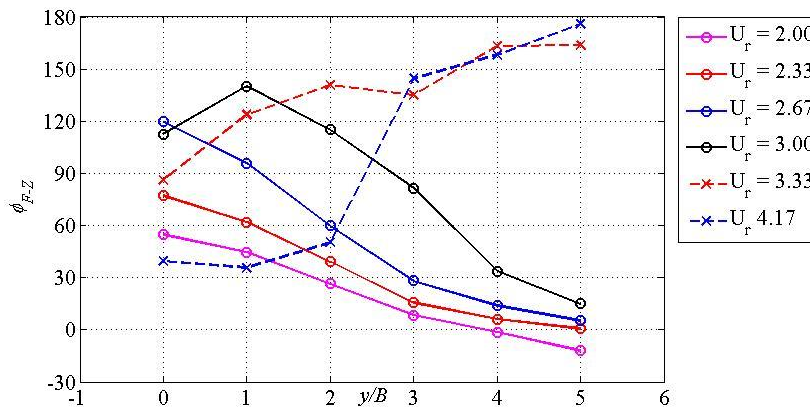


Figure 12. Span-wise variation of the phase angle of the lift force against the displacement as the model experienced the lock-in.

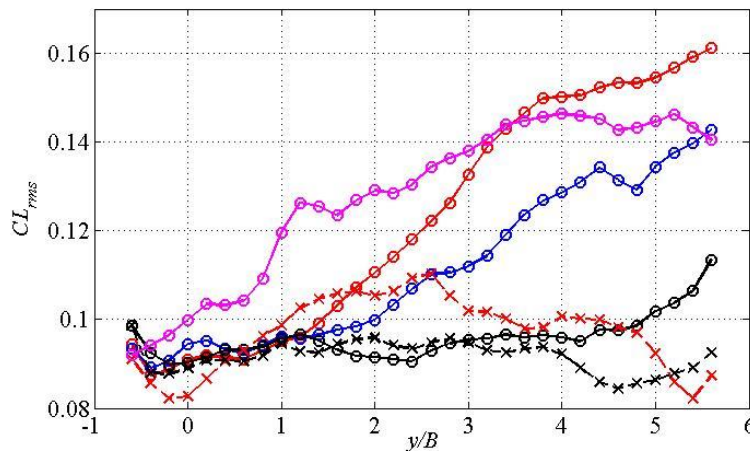


Figure 13. Span-wise variation of C_L' as the model experienced the lock-in; notations are in Figure 12.

4. CONCLUSIONS

Together with the on-going wind tunnel study, the heaving simulation in this computational study has shown the VIV of the 5:1 rectangular cylinder occurred due to two separate mechanisms. The motion-induced leading edge vortex was responsible as a triggering mechanism of the lock-in; the flow feature around the trailing edge where the leading edge vortex impinged on the surface induced a raise in the local suction leading to an increase in the response.

The bending simulation using the flexible model has showed similar behaviour regarding the VIV. The analysis of the surface pressure distribution and the velocity in the wake region has revealed distinct aerodynamics of the flow field in a comparison against the heaving model including a span-wise variation in the pressure distribution and wake structure. Further analysis also showed that, in this case, the VIV lock-in of the flexible model could be originated by some flow features around the portion between $y/B = 1$ and 3, not exactly at the moving end where the maximum displacement was expected to be observed. As the amplitude was built-up, increased pressure gradient in the span-wise direction could potentially diffuse or shift this flow feature towards to the moving end leading to an increase in the response.

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